

SYSTEMS

All systems are designed with reference to maximum efficiency at lowest weight-space penalty.

Propulsion:

Wright J65-W-4 engine
Thrust: 7700 lbs. (military)
Fuel: JP-4 or equivalent

Electrical:

AC POWER—Constant Speed Generator, 3 phase 115/220 volt, 380/440 cps. kva.

AC EMERGENCY POWER—Generator, 115/200 volt, 360/440 cps., 1.5 kva.

DC POWER — Transformer Rectifier, 28 volts, 12 amp.

DC AUXILIARY POWER—Transformer Rectifier, 200 volt AC to 28 volt DC at 55 amp.

Flight Control:

Conventional control surfaces are actuated by the control stick through a dual hydraulic system and rudder pedals. The stabilizer is adjustable and is electrically operated by the pilot to maintain trim through the normal flight range. Manual control of the system is provided for emergencies.

In-Flight Fueling

Removable probe for in-flight fueling is available.

Fuel:

Fuel is carried in an integral wing tank and an internal fuselage tank. For long-range missions additional fuel may be carried in external droppable tanks. In flight fueling capabilities are provided.

Lightweight Ejection Seat
Liquid Oxygen System
Air Conditioning Unit

